CLAIMS LISTING:

1. (Currently Amended) A method for manufacturing a stator or rotor component (21) which is intended during operation to conduct a gas flow, comprising:

providing a first wall part (1, 14, 15, 114) having one edge (3) bearing ~~against the~~ against a flat side (4) of a second wall part (4, 9, 109), extending in the intended radial direction of the component, in such a way that the first wall part extends in the intended circumferential direction of the component, and in that the edge of the first wall part is then laser-welded to the second wall part from an, in the circumferential direction, opposite side of the second wall part in relation to the first wall part in such a way that the joined-together portions of the wall parts form a T-shaped joint (5) and wherein the first wall part (1, 14, 15, 114) is arranged such that it also extends in an intended axial direction of the component.

2. (Original) The method as recited in claim 1, wherein the first wall part (1, 14, 15, 114) is placed essentially perpendicular to the flat side of the second wall part (4, 9, 109).

3. (Original) The method as recited in claim 1, wherein the second wall part (4, 9, 109) is arranged such that it also extends in the intended axial direction of the component.

4. (Cancelled)

5. (Original) The method as recited in claim 1, wherein the second wall part (4, 9, 109), extending in the radial direction, is arranged so as to limit a gas duct (20) in the circumferential direction of the component.

6. (Original) The method as recited in claim 1, wherein the second wall part (4, 9, 109) is arranged such that it has the essentially radial widening for guidance of the gas flow and/or transmission of load during operation of the component.

7. (Currently Amended) ~~The method as recited in claim 1,~~

A method for manufacturing a stator or rotor component (21) which is intended during operation to conduct a gas flow, comprising:

providing a first wall part (1, 14, 15, 114) having one edge (3) bearing against the flat side (4) of a second wall part (4, 9, 109), extending in the intended radial direction of the component, in such a way that the first wall part extends in the intended circumferential direction of the component, and in that the edge of the first wall part is then laser-welded to the second wall part from an, in the circumferential direction, opposite side of the second wall part in relation to the first wall part in such a way that the joined-together portions of the wall parts form a T-shaped joint (5) and wherein the first wall part (1, 14, 15, 114), extending in the circumferential direction, is arranged so as to limit a gas duct (20) in the radial direction.

8. (Currently Amended) ~~The method as recited in claim 1,~~

A method for manufacturing a stator or rotor component (21) which is intended during operation to conduct a gas flow, comprising:

providing a first wall part (1, 14, 15, 114) having one edge (3) bearing against a flat side (4) of a second wall part (4, 9, 109), extending in the intended radial direction of the component, in such a way that the first wall part extends in the intended circumferential direction of the component, and in that the edge of the first wall part is then laser-welded to the second wall part from an, in the circumferential direction, opposite side of the second wall part in relation to the first wall part in such a way that the joined-together portions of the wall parts form a T-shaped joint (5) and wherein the first wall part (1, 14, 15, 114) has a shape which curves essentially in the circumferential direction.

9. (Original) The method as recited in claim 1, wherein the first wall part (14, 15) is placed with a second edge, which is opposite to the first-named edge, bearing against the flat side of a further second wall part (10, 110), which is arranged at a distance in the circumferential direction from the first-named second wall part (9), and is connected thereto.

10. (Original) The method as recited in claim 9, wherein the edge of the first wall part (14, 15, 114) is also laser-welded to this further second wall part (10, 110) from an, in the circumferential direction, opposite side of the second wall part in relation to the first wall part in such a way that the joined-together portions of the wall parts form a T-shaped joint (5).

11. (Original) The method as recited in claim 9, wherein the two wall parts (9, 10, 109, 110) which are spaced apart in the circumferential direction constitute at least part of two different blades or stays for guidance of a gas flow and/or transmission of load.

12. (Original) The method as recited in claim 9, wherein the two second wall parts (9, 10) are formed by a single, substantially U-shaped element (6).

13. (Original) The method as recited in claim 1, wherein the first and second wall part (9, 10, 14, 15) are arranged between an, in the radial direction, inner and outer ring element (7, 8).

14. (Original) The method as recited in claim 13, wherein the second wall part (9, 10) is connected to at least one of the ring elements (7, 8) by laser-welding from an, in the radial direction, opposite side of the ring element in relation to the second wall part in such a way that the joined-together portions form a T-shaped joint (5).

15. (Original) The method as recited in claim 12, wherein the first and second wall part (9, 10, 14, 15) are arranged between an, in the radial direction, inner and outer ring element (7, 8) and the U-shaped element (6), prior to the laser-welding of the wall parts, is arranged between the inner ring element (7) and the outer ring element (8).

16. (Original) The method as recited in claim 1, wherein the stator or rotor component (21, 23) has an essentially circular cross-sectional shape and in that a plurality of ducts (20) for conduction of the gas flow extend in the axial direction between an inner and an outer ring.

17. (Original) The method as recited in claim 1, wherein the stator or rotor component (21, 23) is intended for a gas turbine.

18. (Original) The method as recited in claim 1, wherein the stator or rotor component (21, 23) is intended for a jet engine.